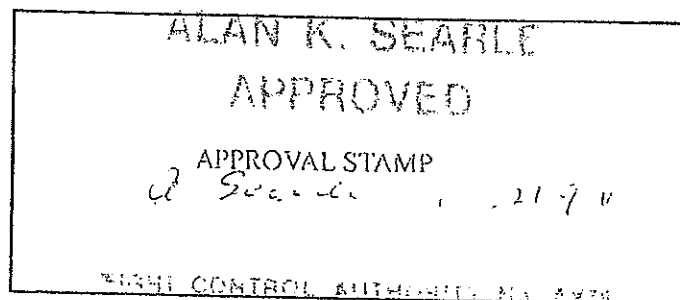


LOAD DATA SHEET

AIRWEIGH AVIATION SERVICES
P.O. Box 2340, Ivanhoe East, Vic 3079. Phone 0418 166 700.

Aircraft Type: **Piper PA-28-161.** Aircraft Registration : **VH-ZFX.**
Approved Loading System : Piper PA-28-161 Pilots Operating Handbook (POH). Sn: 2842266

AUTHORISED	DATE	DATE OF EXPIRY	ISSUE
A.K. SEARLE AV 70	21-9-2011.	Indefinite subject to CAO 100.7	One.



AIRCRAFT WEIGHT AND CENTRE OF GRAVITY DATA

ITEM	WEIGHT	ARM	MOMENTS	CONFIG
EMPTY	717.3 kg.	2194 mm.	1,573,756 kg.mm.	4 Seats
EMPTY	1581.3 lb.	86.377 inches.	136,587,545 lb.in.	4 Seats.
DATUM: 1991 mm. Fwd WLE @ Intersection of straight and taper section.				

Notes : Empty Weight includes Unusable Fuel and Full Engine Oil.

It is the responsibility of the Pilot and the Operator to ensure that the aircraft is loaded correctly. Load in accordance with the limitations defined in the approved Flight Manual.

Weight data validated on the basis of running totals in lieu of weighing.

SECTION 6
WEIGHT AND BALANCE

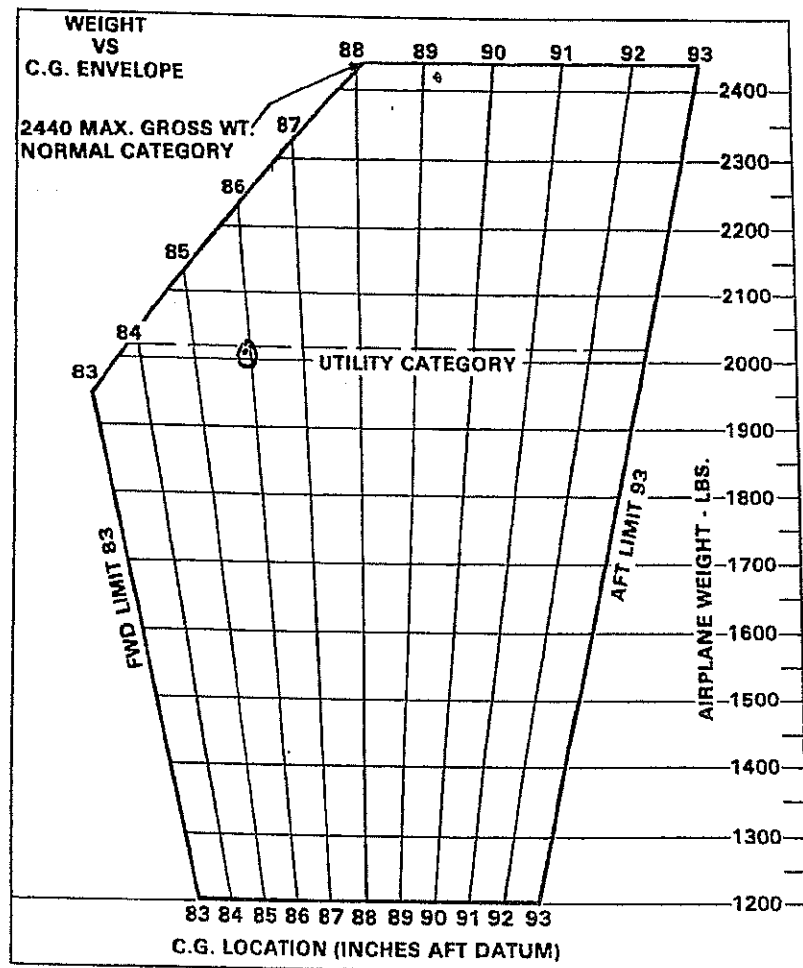
PA-28-161, WARRIOR III

	Weight (Lbs)	Arm Aft Datum (Inches)	Moment (In-Lbs)
Basic Empty Weight			
Pilot and Front Passenger		80.5	
Passengers (Rear Seats)*		118.1	
Fuel (48 Gallon Maximum)		95.0	
Baggage* (200 Lbs. Maximum)		142.8	
Ramp Weight (2447 Lbs. Normal, 2027 Lbs. Utility Maximum)			
Fuel Allowance For Engine Start, Taxi and Run Up	-7	95.0	-665
Takeoff Weight (2440 Lbs. Normal, 2020 Lbs. Utility Maximum)			

Totals must be within approved weight and C.G. limits. It is the responsibility of the airplane owner and the pilot to insure that the airplane is loaded properly. The Basic Empty Weight C.G. is noted on the Weight and Balance Data Form (Figure 6-5). If the airplane has been altered, refer to the Weight and Balance Record for this information.

*Utility Category Operation - No baggage or aft passengers allowed.

WEIGHT AND BALANCE LOADING FORM
Figure 6-11



C.G. RANGE AND WEIGHT
Figure 6-15

LOAD CHART BC/PA161/1 PIPER PA28-161 WARRIOR

*APPLICABLE TO ALL PA28-161
WARRIOR II AND III AIRCRAFT
S/N 28-7716001 AND ON.

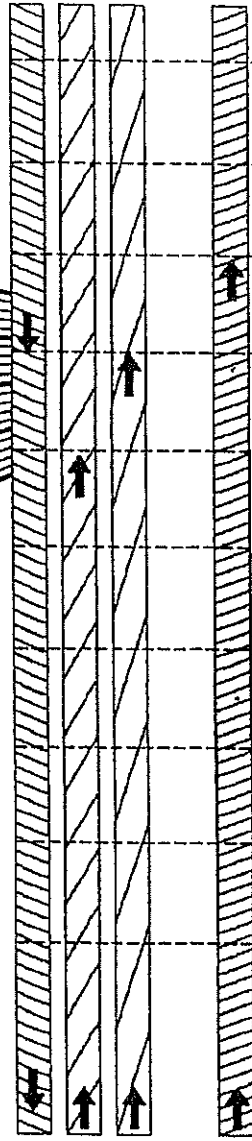
Aeroweigh Ltd.
BRUCE GLISSOLD
AUTHORITY NUMBER AN-9
PHONE 9791 5267 A.H. 8645 1482
MOBILE 018 265 622

BASIC ARM (mm)
2100 2200 2300

50 Kg PER DIVISION

BASIC WEIGHT	
PILOT/CO-PILOT	
REAR SEATS	
BAGGS (MAX 90.7)	
ZERO FUEL WEIGHT	
FUEL (MAX 130)	
TAKE-OFF WEIGHT	
FUEL BURN-OFF	
LANDING WEIGHT	

BASIC WEIGHT (Kg)

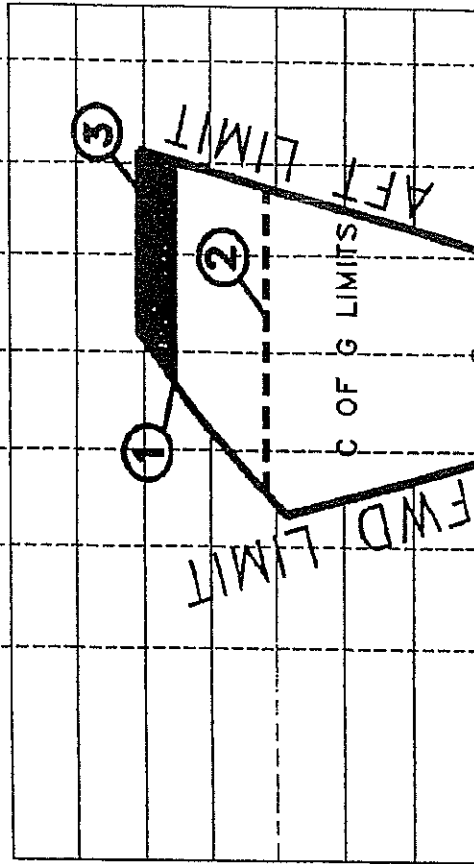


GROSS WEIGHT (kg)
1200
1100
1000
900
800
700

MAXIMUM TAKE-OFF WEIGHT

- ① NORMAL — 1055 Kg
- ② UTILITY - - - 916 Kg
- ③ NORMAL — 1107 Kg

☐	☐	BAGGS
☐	☐	



③ OPERATION AT WEIGHTS IN EXCESS OF 1055 KG IS ONLY
APPLICABLE TO AIRCRAFT SERIAL 28-8316001 AND UP
OR ANY AIRCRAFT MODIFIED TO S.T.C. SA00397NY

26X

PA-28-161, WARRIOR III

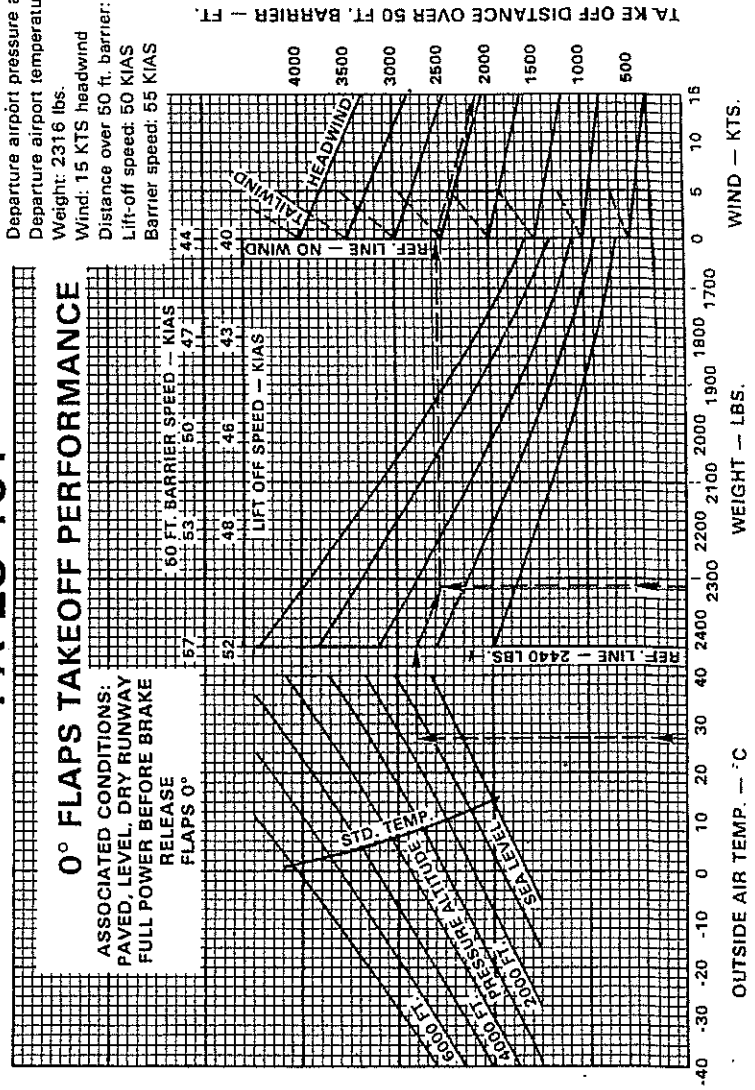
SECTION 5
PERFORMANCE

PA-28-161

Example:
Departure airport pressure altitude: 1500 ft
Departure airport temperature: 27°C
Weight: 2316 lbs.
Wind: 15 KTS headwind
Distance over 50 ft. barrier: 2100 ft
Lift-off speed: 50 KIAS
Barrier speed: 55 KIAS

0° FLAPS TAKEOFF PERFORMANCE

ASSOCIATED CONDITIONS:
PAVED, LEVEL, DRY RUNWAY
FULL POWER BEFORE BRAKE
RELEASE
FLAPS 0°

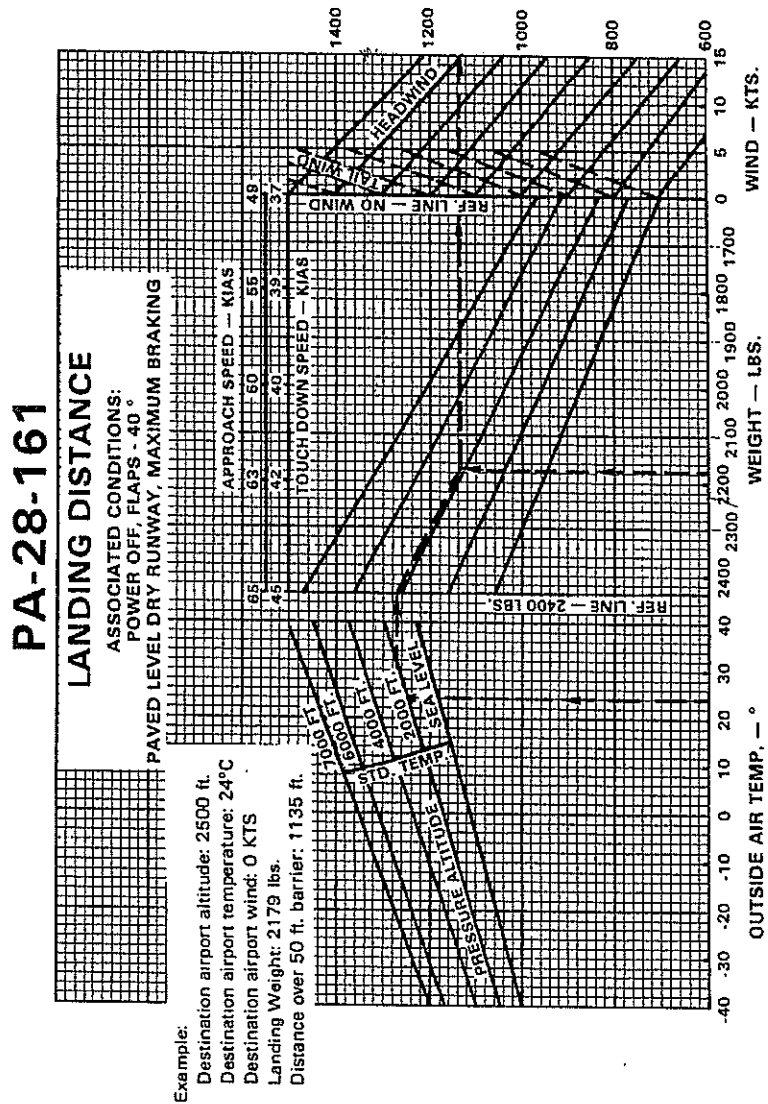


0° FLAPS TAKEOFF PERFORMANCE

Figure 5-9

ISSUED: JULY 12, 1995

REPORT: VB-1610
5-15



LANDING DISTANCE

Figure 5-35

Climb Gradients

1 Take off climb performance (20.7.4 / 7.1)

In the take-off configuration with landing gear extended, an aeroplane must have the ability to achieve a climb gradient of 6% at takeoff safety speed (TOSS), without ground effect, and with all engines operating at take-off power.

2 En-route climb performance (20.7.4 / 8.3)

Single-engined aeroplanes must have the ability to climb at a gradient of 4.5% at an airspeed not less than 1.2Vs at all heights up to 5000ft in standard atmospheric conditions with the engine operating at maximum continuous power, undercarriage (if retractable) and flaps retracted.

3 Landing Climb performance (20.7.4 / 9.1)

In the landing configuration with all engines operating at take-off power an aeroplane must have the ability to climb at a gradient of 3.2% in standard atmospheric conditions at a speed not exceeding 1.3Vs

Formula

$$\text{RoC} = ((\text{GS} \times 6080) / 60) \times (\text{CG} / 100)$$

Or

$$\text{CG} = \frac{\text{ROC}}{(\text{G.S} \times 6080) / 60}$$